

Rapid design method and new contours for a class of three dimensional supersonic nozzle of arbitrary exit cross section

Toufik Zebbiche*

Institute of Aeronautics and Space Studies, University of Blida 1, BP 270, Blida 09000, Algeria

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Abstract. The aim of this work is to develop a new and rapid numerical method for designing a new contour of the supersonic nozzle with arbitrary exit cross sections as a class of three dimensional nozzle extracted from the calculation of stationary flow solutions in axisymmetric nozzle. The application is made for nozzle giving a uniform and parallel flow at the exit section. The determination of the points of the axisymmetric nozzle contours in a non-dimensional manner with respect to the arbitrary throat radius is necessary. The exit section of the nozzle must be discretized at several points. The radii of the points of the throat section are determined by equalizing the ratio of the axisymmetric critical sections corresponding to each selected point of the exit section. Each point passes a contour of the nozzle to the throat where their position is determined by the multiplication of the non-dimensional positions of the axisymmetric nozzle point by the throat radius of this contour. A uniform portion is added at the end of each contour to compensate the decrease in its length. Longitudinal discretization of the nozzle is necessary. The flow properties of each point are the same as those of the points of the axisymmetric nozzle. The temperature and the deviation of the flow at each point of any section are determined by their interpolations between two successive points of each contour. The Mach number, the pressure and the density are determined accordingly. The application is made for air at high temperature lower than the dissociation threshold of the molecules.

Keywords: Supersonic flow / supersonic axisymmetric minimum length nozzle / high temperature / calorically imperfect gas / method of characteristics / specific heat at constant pressure / error

1 Introduction

Numerical computing obliged us to think of developing efficient methods that meet the demands of convergence with high precision and in a smallest possible computer execution time. Generally the execution time for design problems is quite high despite the power of calculator used since there is repetition of calculation. The methods developed for determining the contours of the three dimensional nozzles are based on the determination of the internal stream lines using the definition of the stream function [1–6]. As the flow is considered as perfect and non-viscous, the stream lines can represent rigid surfaces and hence a wall shape of obstacles [7]. In our case these stream lines can represent the wall of the nozzle [1–6]. The use of these methods requires a very high computational time since the search of the points of the stream lines is based on the use of the segments in upward and downward characteristics from the developed internal mesh in the axisymmetric nozzle, which becomes impossible for fine

meshes, in a time when a fine mesh give a high precision. Theoretically the fine meshes gives a good accuracy for the design, but in parallel, it falls failing for the computer design of the three dimensional nozzles based on the stream lines search.

The segment number of the internal mesh characteristics of the axisymmetric nozzle depends on many of the descending characteristics in the Kernel region and the step used in the transition region. Generally the number of points found on the wall is much greater than the number of descending characteristics in the Kernel region. We are interested in axisymmetric nozzles giving a uniform and parallel flow to the exit section [8–11]. The number of information on the flow parameters ($x, y, P, T, \rho, M, \theta$ and ψ) at each internal mesh point becomes very large, which requires a very large storage reservation in the computer memory, and which gives in parallel a very important calculation time for the design of the three dimensional and axisymmetric nozzles. For a small number of N_S and N_P , the design becomes possible, but with moderate precision. This gives that the methods developed in references [1–6] are limited by a certain maximum of discretization. It is not necessary, in the general case, to know the properties of the

* e-mail: z_toufik270169@yahoo.fr

flow inside the nozzle. Only the axisymmetric nozzle contour is desired. But if the design is made as a passage for the design of the three dimensional nozzle, it is necessary to determine the internal mesh information according to references [1–4].

According to [1], the design is made only for the *PG* model case, calorically and thermally perfect from the design of the axisymmetric nozzle using the internal mesh for the search of the stream lines points.

The reference [2] deals with the design of the three dimensional nozzles from a given axisymmetric shape taken as a second-degree polynomial for the use of the *PG* model, using the definition of a stream line always in an internal mesh. References [3–4] deal with the problem presented in reference [2] but only for a square section. In addition a truncation is presented in the reference [4].

In reference [5], the author made the design of the nozzles 3D of arbitrary exit section by the method of the characteristics in the case of *PG* model.

In the reference [6], the authors made the design of the nozzles 3D from the axisymmetric nozzle in the context of the *HT* model using the axisymmetric nozzle design and the search of stream lines from the internal mesh.

In references [1–5], the *PG* model is used for the design. Not to mention the effectiveness of the method, this model gives good results if the M_E does not exceed about 2.00, and T_0 does not exceed the 1000 K. approximately [6,8,11,12]. The *HT* model is used for the same design problem in view of the presentation of the corrections to the results obtained in the references [1–5] in the case where M_E exceeds 2.00 and T_0 exceeds 1000 K. By way of information, the *HT* model presents a generalization of the *PG* model.

References [13–16] have studied and analyzed analytically and numerically the supersonic flow by determining the parameters of flow through the Laval nozzle, as well as CFD validation. The study is done for the two-dimensional nozzle using *PG* model already discussed. In reference [16], the authors studied the outside adaptation of the nozzle with consideration of the shock birth and a separation of the flow.

Regarding the design of the nozzle 3D according to references [1–6], for N_T and N_L points chosen, one needs to treat $N_T \times N_L \times N_S$ segments. It is true that a technique is normally opted not to go through the N_S segments for even point of the nozzle 3D. But as even, the number of segments to be processed is very large. It can be said that the method developed in references [1–6] is practically limited by a maximum of discretization of the axisymmetric nozzle, at a time when the high discretization is preferred in order to obtain very good results.

The aim of this work is to solve the disadvantages found in the methods used for the design of the nozzles 3D presented in references [1–6], by developing a new very fast and efficient numerical method that can design the new contours of the nozzles 3D of arbitrary exit section in the *HT* model as a special class of Laval nozzle, only from the boundary axisymmetric nozzle wall, without going through the use of the stream function and storage of the internal meshing information, to derive the properties of the flow at all points of the wall of the nozzle 3D. The first problem is to design the axisymmetric nozzle in dimensionless values with

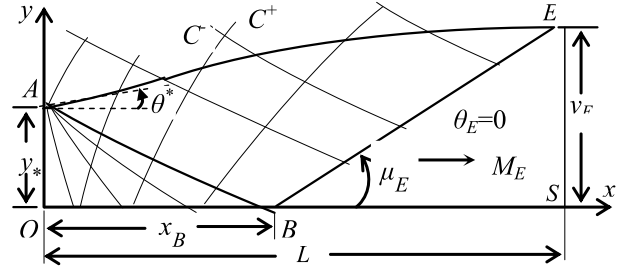


Fig. 1. Presentation of the axisymétric *MLN* flow field.

respect to the throat radius, that is, determine the positions of the wall in terms of $(x/y_*$ and y/y_*). The axisymmetric *MLN* is chosen in our work seen its practical and current use in aerospace applications of missiles and satellites launchers because it has better performances compared to other types of supersonic nozzles in addition to its simplicity of construction [8,11–17]. The application is made for the case of air with *HT*, lower than the dissociation threshold of the molecules. The flow is considered as adiabatic, irrotational and stationary.

2 Axisymmetric *MLN* design contour

We are interested in the *MLN* giving a uniform and parallel flow at the exit section due to elimination of the thrust losses caused by the inclination of the wall and to have a complete expansion through the nozzle. The general shape of the *MLN* as well as the various zones of the flow are shown in Figure 1. The contour search (x, y) of the nozzle wall as well as the flow properties $(M, T, \theta, P$ and $\rho)$ at all flow internal points and the boundary points are done simultaneously and by the use of the *HT MOC* [8]. The application is made for air, lower than the dissociation threshold of the molecules.

The design results of the axisymmetric nozzle depend on the mesh quality used, which is itself dependent on two important parameters which are $\Delta\theta$ and Δ characterizing respectively the refinement and the mesh quality in the Kernel and transition zones. The internal mesh therefore contains N_S total segments of the descending rising characteristics and N_P internal points. The wall of the nozzle then contains N_F points. The values of N_F, N_S, N_P and ε depend essentially on $\Delta\theta, \Delta, M_E$ and T_0 .

The equations for axisymmetric supersonic non viscous flow of perfect gas with *HT* by the *MOC* can be summarized by:

$$\begin{cases} -\frac{C_P(T)}{2H(T)}\sqrt{M^2(T)-1}dT + d\theta = \frac{\sin\theta\sin\mu}{y\cos(\theta-\mu)}dx \\ \frac{dy}{dx} = tg(\theta-\mu) \end{cases}, \quad (1)$$

$$\begin{cases} -\frac{C_P(T)}{2H(T)}\sqrt{M^2(T)-1}dT - d\theta = \frac{\sin\theta\sin\mu}{y\sin(\theta+\mu)}dx \\ \frac{dy}{dx} = tg(\theta+\mu) \end{cases}. \quad (2)$$

With

$$H(T) = \int_T^{T_0} C_P(T) dT. \quad (3)$$

The M and T at HT are connected by [18,19]:

$$M^2(T) = \frac{2H(T)}{\gamma(T)RT}. \quad (4)$$

With

$$\gamma(T) = \frac{C_P(T)}{C_P(T) - R}, \quad (5)$$

$$\mu = \arcsin\left(\frac{1}{M}\right). \quad (6)$$

The ratios ρ/ρ_0 and P/P_0 are calculated by [18,19]:

$$\frac{\rho}{\rho_0} = \exp\left(-\int_T^{T_0} \frac{C_P(T)}{\gamma(T)RT} dT\right), \quad (7)$$

$$\frac{P}{P_0} = \frac{T}{T_0} \frac{\rho}{\rho_0}. \quad (8)$$

As the flow through the throat and the exit section is unidirectional, the theoretical ratio of critical sections at HT , given by equation (9) remains valid for the convergence of the numerical found results [18,19]:

$$\begin{aligned} \left(\frac{y_E}{y_*}\right)_{Exact}^2 &= \frac{A_E}{A_*} \\ &= \exp\left(\int_{T_E}^{T_*} \left[\frac{C_P(T)}{\gamma(T)RT} - \frac{C_P(T)}{2H(T)}\right] dT\right). \end{aligned} \quad (9)$$

To validate the numerical results of design, one can calculate the relative error committed the ratio of the critical sections by the following relation:

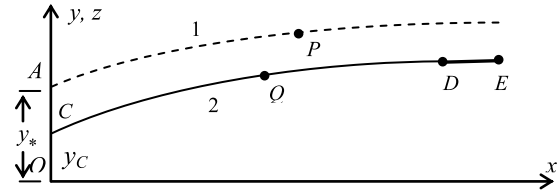
$$\varepsilon_{(y_E/y_*)}(\%) = \left| 1 - \frac{(y_E/y_*)_{Computed}}{(y_E/y_*)_{Exact}} \right| \times 100. \quad (10)$$

The value of $(y_E/y_*)_{Computed}$ illustrated in relation (10) represents the dimensionless exit radius determined according to the chosen problem discretization.

For air $R = 287.1029 \text{ J}/(\text{kg K})$. For the PG model, the value of $\gamma = 1.402$. The range of the variation of M_E is up to 5.00 and for T_0 is up to 3500 K according to the references [8,11,18–20] so as not to have dissociation of the molecules.

3 Presentation of the developed method

A new very rapid and effective method is proposed for designing the novel contours of the nozzle 3D from the shape of the axisymmetric nozzle without going through the search of the stream lines and the properties of the flow in the internal mesh of the axisymmetric nozzle. This time, we are only interested in the contour properties of the



1 : Axisymmetric main nozzle
2 : Intermediate contour corresponding to y_C .
DE : Straight horizontal contour of the nozzle 3D.

Fig. 2. Intermediates contour of the axisymmetric nozzle for nozzle 3D.

axisymmetric nozzle wall. Then during the design of the axisymmetric nozzle, the properties of the internal flow mesh are omitted and not stored, which make in this step the fairly quick calculation process which is a reported advantage in this first step. During the design of the axisymmetric nozzle, the radius of the throat is chosen in the non-dimensional form. The coordinates of the contour points of the nozzle wall are calculated by $(x/y_*, y/y_*)$. Consequently, the length and the exit radius of the nozzle are normalized again by y_* . The values L/y_* and y_E/y_* respectively are obtained. The choice of the exit section shape of the nozzle 3D is arbitrary, which must be discretized at several points and placed in the exit section of the axisymmetric nozzle. The radii y_C of the points of the critical section, which the latter has the same shape as the exit section, are determined by equalizing the ratio of the corresponding axisymmetric critical sections to each selected point of the exit section. Each point of the exit section passes a contour of the nozzle 3D to the throat.

For a given radius of the throat $0 < y_C < y_*$ of the axisymmetric nozzle, it is possible to determine the position of the forming points of a contour which can represent a wall of an intermediate nozzle of the nozzle 3D whose position of these points is determined by multiplying the non-dimensional positions of the axisymmetric nozzle by the throat radius of this contour. Then the position of the new point Q , see Figure 2, is determined from the position of point P by the following relations:

$$x_Q = \left(\frac{x_P}{y_*}\right)y_C, y_Q = \left(\frac{y_P}{y_*}\right)y_C. \quad (11)$$

The flow properties (M , P/P_0 , T/T_0 , ρ/ρ_0 and θ) at this new point Q of this contour are the same as those of the point P of the axisymmetric nozzle.

By way of information, the reference mark chosen on the section of the nozzle is named by Oyz . Then the axis of the nozzle is the Ox .

The intermediate contour which can also represent the wall of the nozzle 3D is included in the main axisymmetric shape as shown in Figure 2. The longitudinal length of this contour is less than the main length of the axisymmetric nozzle. It is given by:

$$L_2(y_C) = \left(\frac{L_1}{y_*}\right)y_C. \quad (12)$$

The value given by relation (12) represents the length of the intermediate contour between points C and D of Figure 2. A uniform portion is added at the end of each contour to compensate the decrease in its length.

By varying the value of y_C , we can find other contours we will use to form the nozzle 3D.

In the first place, it is necessary to arbitrarily choose the shape of the exit section of the nozzle 3D which is to be placed in the exit section of the axisymmetric nozzle as shown in Figure 3. The discretization of this shape in an N_T number of points and the $(y, z, r$ and $\beta)$ as well as $(M, T/T_0, P/P_0, \rho/\rho_0$ and $\theta)$ in all these points is necessary. The flow properties are the same for all points of the exit section of the axisymmetric nozzle and are equal to the values corresponding to $M = M_E$.

For our method it is not necessary to calculate the value of the stream function at each point as it is used in references [1,6], so a slight reduction in calculation time for our method, given the large number of mathematical operations won.

A particle which runs through the flow field on this contour from point C will arrive with an exit Mach number M_E at the point D . The portion of this contour between the point D and the point E of Figure 2 is added since the contour ends at point E because the exit section of the nozzle 3D is placed exactly at the exit section of the axisymmetric nozzle. This contour portion is horizontal since M_E is reached at point C . The flow properties on this segment are the same and they are equal to those of the exit section of the nozzle.

So we are only interested in the physical properties at the points of this contour. The problem therefore becomes the determination of the ordinate $y_C < y^*$ which will identify this contour and which verifies the equality of the ratios of the axisymmetric critical sections between the axisymmetric main nozzle and that of the contour which forms the nozzle 3D. Since the contour passes through the Q points of the exit section as shown in Figure 3, the equality of the ratios of the axisymmetric sections gives the following result:

$$y_C^2 = \frac{z_Q^2 + y_Q^2}{\left(\frac{A_E}{A_*}\right)}. \quad (13)$$

For each point Q of the exit section of Figure 3, there is a contour which passes through this point and passes through the longitudinal length of the nozzle to the throat and cuts the throat at a point having a radius $y_C < y^*$ found by the relation (13). The number of points found on each intermediate contour is the same and it is equal to that of the axisymmetric main nozzle plus 1, since point E has been added according to Figure 2. Each contour forms a plane of the nozzle inclined by an angle β as shown in Figure 3. Then all the points of the intermediate contour have the same value of the angle β .

4 Flow properties in a vertical section

The points found on each contour are not placed exactly on straight sections. It is highly recommended to determine

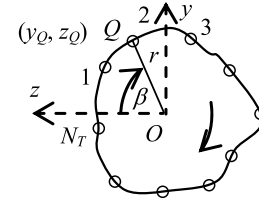


Fig. 3. Transverse discretization of the exit section of the nozzle 3D.

the shape of the nozzle in each section by determining the positions (y, z) and the properties $(M, P/P_0, T/T_0, \rho/\rho_0$ and $\theta)$ on each point of a section.

The discretization of the longitudinal direction of the nozzle in N_L section must be presented as illustrated in Figure 4. In this Figure, only the abscissa x_Q of the N_T points of each section is known.

The point Q to be searched for the nozzle 3D is located on a limited segment between two successive points of corresponding contour of the nozzle 3D according to Figure 4. The segment containing the point Q of the nozzle 3D wall must verify the following condition:

$$(x_L - x_Q) \times (x_R - x_Q) \leq 0. \quad (14)$$

The coordinates (y_Q, z_Q) of this point can be determined by:

$$y_Q = r \cos(\beta), \quad (15)$$

$$z_Q = r \sin(\beta). \quad (16)$$

The value of β is already determined at each point of the exit section chosen in the first step. It is the same at all points of the corresponding contour. It can be determined by the following relation:

$$\beta = \text{atan}\left(\frac{y_Q}{z_Q}\right). \quad (17)$$

The point Q in relation (17) is that of the exit section and not those given by relations (15) and (16). The point Q given by (15) and (16) and that used in relation (17) lie on the same longitudinal plane formed by the contour under consideration but are not merged.

The value of r can be determined by the following relationship, referring to Figure 3.

$$r = \frac{(r_R - r_L)}{(x_R - x_L)}(x_Q - x_L) + r_L. \quad (18)$$

With

$$r_L = \sqrt{y_L^2 + z_L^2}, \quad (19)$$

$$r_R = \sqrt{y_R^2 + z_R^2}. \quad (20)$$

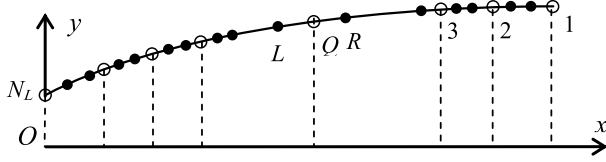


Fig. 4. Longitudinal discretization of the nozzle 3D.

The point Q in relation (14) represents all the points of the Figure 3. The properties at points L and R of Figure 4 represent those of intermediate contour already calculated. It can also be considered that the Figure 3 also represents the shape of the section of the nozzle 3D in each longitudinal station.

The coordinates (x, y, z) and the flow properties $(M, T/T_0, P/P_0, \rho/\rho_0)$ and θ at each point of the section can be determined for each contour of the nozzle 3D by starting the calculation from the exit section towards the throat of the nozzle. The abscissa of these points belonging to a section of the nozzle of Figure 4 is given by x_Q chosen arbitrarily. The points in bold circles in Figure 4 represent the intermediate contour determined from the main axisymmetric shape and corresponding to the throat radius y_C . The points in hollow circles illustrated in Figure 4 represent the points at the selected sections of the nozzle 3D, where it is desired to determine their shapes and the physical properties.

The computation of the angle β by computer from relation (17) generally gives the angle in the interval $[-\pi/2, \pi/2]$. Then we have a cosine and sinus calculation problem of the four coordinates (y, z) and $(-y, z)$, $(y, -z)$ and $(-y, -z)$ from relations (15) and (16).

The first and the fourth points have the same ratio y/z . Then the calculation of the angle by the relation (17) gives the angle corresponding to the point of the first quadrant. Then for the point of the third quadrant (fourth point), one will have a problem of calculation of the corresponding angle. To solve this problem, the angle of the third quadrant must be corrected by $2\pi - \beta$ where β is the angle for the point of the first quadrant.

For the second and third points we will have the same problem. The computation by computer will confuse the calculation of the angle β for these two points by the relation (17). Then the angle β of the second quadrant point $(y, -z)$ is equal to $\pi +$ the angle β of the point $(-y, z)$ of the fourth quadrant.

The second problem is found for the particular points $(y=0.0, z)$ and $(y=0.0, -z)$ of the horizontal axis. The angle for the first point is $\beta = 0.0$ (no problem for this point) and the second is $\beta = \pi$. The computation by computer of the angle β corresponding to the point of the negative axis by the relation (17) still gives $\beta = 0.0$. Then we must correct this result by adding π for this point.

The third problem encountered is the computation of the angle β by computer for the particular points $(y, z=0.0)$ and $(-y, z=0.0)$ of the vertical axis. We will have a division by zero if we use the relation (17). In this case, the calculation is ignored by the relation (17) by assignment $\beta = \pi/2$ for the first point on the positive axis and $\beta = -\pi/2$ for the second point on the negative axis.

In this way it is possible to calculate exactly the corresponding angle β for all points of Figure 3 according to its position and there will be no cosine and sinus calculation problem in relation (15) and (16). Therefore, the position of the corresponding point is exactly calculated.

The flow parameters at this new point Q of the Figure 4 can be determined by first interpolating the parameters T and θ by the following relations:

$$T_Q = \frac{S_{LQ}}{S_{LR}}(T_R - T_L) + T_L, \quad (21)$$

$$\theta_Q = \frac{S_{LQ}}{S_{LR}}(\theta_R - \theta_L) + \theta_L. \quad (22)$$

With

$$S_{LQ} = \sqrt{(y_L - y_Q)^2 + (z_L - z_Q)^2}, \quad (23)$$

$$S_{LR} = \sqrt{(y_L - y_R)^2 + (z_L - z_R)^2}. \quad (24)$$

The ratio T/T_0 at the point Q is equal to T_Q/T_0 . The values of M , ρ/ρ_0 and P/P_0 at the point Q can be determined by relations (4), (7) and (8) respectively, by replacing T by T_Q that obtained by the relation (21). The integration of the relation (8) is done by the Gauss Legendre quadrature [21,22] for the speed of calculation with high accuracy.

By doing the same procedure for all the N_T points of the selected section of the nozzle 3D of Figure 3 in all the N_L stations of Figure 4, it is possible to determine the positions of all the points of the selected sections of the nozzle 3D.

5 Comparison and validation

The validation of the developed method is controlled by choosing a circular section of the nozzle 3D and calculating the corresponding complete shape of the nozzle 3D at each longitudinal station by applying the developed method and comparing it with the results of the axisymmetric nozzle. The shape of the nozzle in all longitudinal sections must be exactly circular in shape. To confirm the circular shape, we calculate the vector radius of all the points forming a section. This radius must be the same in all respects and independently of the angle β . By repeating the process for all selected sections of the nozzle, it can be confirmed that the nozzle is of circular cross-section. Hence the method is validated. It is possible to widen the validation by calculating the flow properties at each point, and to calculate the corresponding angle β . Then it is necessary to find for each section that the flow parameters do not depend on the angle β . In this case, it can be said that the program realized on the basis of the developed method gives good results for any chosen section.

For an arbitrarily chosen non-circular section, the shape of the section at all longitudinal stations must not necessarily be kept on a scale except with the shape of the exit section and the throat.

Table 1. $\Delta\theta$ effect on the mesh of the axisymmetric nozzle for $\Delta = 0.1$, $M_E = 3.00$ and $T_0 = 2000$ K.

$\Delta\theta$ (deg)	N_F	N_S	N_P	ε (%)
3.0	78	1255	748	2.441
2.0	83	1950	1128	1.858
1.0	88	2519	1472	1.119
0.5	105	5186	2809	0.668
0.1	206	31865	16955	0.216
0.05	338	97074	50275	0.139
0.01	1381	1840539	931426	0.056

Table 2. Δ effect on the mesh of the axisymmetric nozzle for $\Delta\theta = 0.1$ (deg), $M_E = 3.00$ and $T_0 = 2000$ K.

Δ	N_F	N_S	N_P	ε (%)
3.0	142	19321	10683	4.622
2.0	143	19539	10792	3.153
1.0	146	20185	11115	1.168
0.5	153	21495	11770	0.496
0.1	206	31865	16955	0.216
0.05	273	44817	23431	0.204
0.01	809	148519	75282	0.198

Table 3. M_E effect on the mesh of the axisymmetric nozzle for $\Delta\theta = 0.1$ (deg), $\Delta = 0.1$ and $T_0 = 2000$ K.

M_E	N_F	N_S	N_P	ε (%)
1.50	49	1800	1193	0.160
2.00	96	7413	4345	0.210
3.00	206	31865	16955	0.216
4.00	334	67819	36023	0.182
5.00	507	117295	61356	0.139

Table 4. T_0 effect on the mesh of the axisymmetric nozzle for $\Delta\theta = 0.1$ (deg), $\Delta = 0.1$ and $M_E = 3.00$.

T_0 (K)	N_F	N_S	N_P	ε (%)
<i>PG</i>	186	25750	13921	0.232
1000	197	29229	15208	0.216
2000	206	31865	16955	0.216
3000	210	32799	17583	0.217

6 Results and comments

Table 1 shows the effect of $\Delta\theta$ of the Kernel region on the boundary and internal mesh parameters N_F , N_S , N_P and the precision ε for the axisymmetric nozzle when using the *MOC* with $\Delta = 0.1$, $M_E = 3.00$ and $T_0 = 2000$ K.

Table 2 shows the effect of Δ of the transition region on boundary and internal mesh parameters N_F , N_S , N_P and ε obtained for the axisymmetric nozzle by using *MOC* when $\Delta\theta = 0.1$ deg, $M_E = 3.00$ and $T_0 = 2000$ K.

Table 3 shows the effect of M_E on the boundary and internal mesh N_F , N_S , N_P and ε for the axisymmetric nozzle using the *MOC* when $\Delta\theta = 0.1$ deg, $\Delta = 0.1$ and $T_0 = 2000$ K.

Table 4 shows the effect of T_0 on the boundary and internal mesh N_F , N_S , N_P and ε for the axisymmetric nozzle using the *MOC* when $\Delta\theta = 0.1$ deg, $\Delta = 0.1$ and $M_E = 3.00$.

Table 5 represents the values of N_F , N_S , N_P and ε of a case generally giving a very fine mesh in the axisymmetric nozzle. This mesh is obtained when $\Delta\theta = 0.01$ deg, $\Delta = 0.1$ and $M_E = 3.00$ and $T_0 = 2000$ K.

Table 5. Unfavorable case on the axisymmetric nozzle mesh for $\Delta\theta=0.01$ (deg), $\Delta=0.01$, $M_E=5.00$ and $T_0=3500$ K.

N_F	N_S	N_P	ε (%)
5542	13317795	6675864	0.027

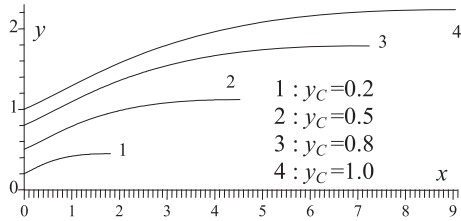


Fig. 5. Intermediate contours of the axisymmetric nozzle without the uniform part *BE*.

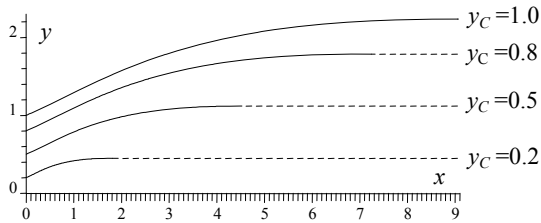


Fig. 6. Intermediate contours of the axisymmetric nozzle with the uniform part *BE*.

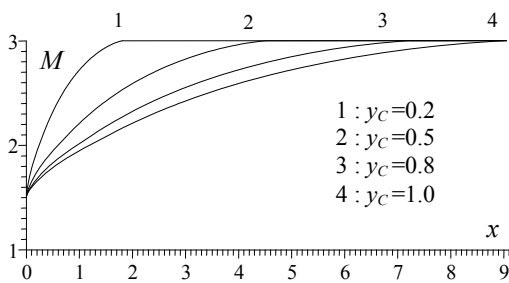


Fig. 7. Variation of M on the contours of [Figure 6](#).

It is noted that the internal mesh N_S , N_P characteristics and the boundary mesh N_F characteristics of an axisymmetric nozzle depend mainly on $\Delta\theta$, Δ , M_E and T_0 . The number N_F of points obtained on the boundary of the wall is very small compared to the numbers N_S of the segments and N_P of internal mesh points in an axisymmetric nozzle. Then the methods presented in the references [1–6] based on the search of the internal stream lines from internal mesh require a very high computer execution time and which becomes impossible for the fine discretization as the example of the [table 5](#). While our present method based on the points N_F of the boundary of the nozzle requires a very small execution time and that becomes negligible before the time of execution of the methods of the references [1–6].

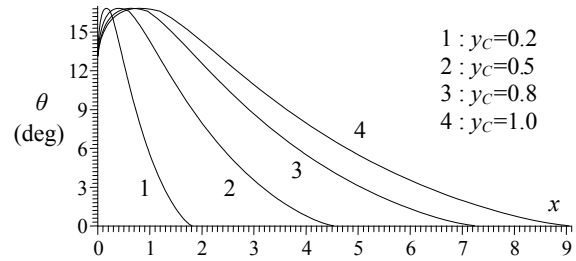


Fig. 8. Variation of θ on the contours of [Figure 6](#).

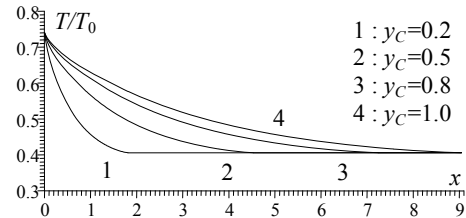


Fig. 9. Variation of T/T_0 on the contours of [Figure 6](#).

According to [table 5](#), when $\Delta\theta=0.01$ deg and $\Delta=0.01$, the accuracy of the results of the axisymmetric nozzle is equal to $\varepsilon=0.027\%$ which is an acceptable error for calculation. But in parallel the methods presented in the references [1–6] use $N_S=13317795$ and $N_P=6675864$ for the search of the profiles of the nozzle 3D. While our method uses only the points of the boundary equal in this case to $N_F=5542$. From the references [1–6], the N_S and N_P numbers are used to search for each $N_L \times N_T$ points of the nozzle 3D. This demonstrates that the method used in those references is limited by a maximum of N_S and N_P corresponding to the precise values of $\Delta\theta$ and Δ of discretization. Given the obtained number N_F , the computer execution time of our method is very small and incomparable to that of the method of [1–6].

[Figures 5](#) and [6](#) represent the intermediate contours of the axisymmetric nozzle wall, respectively without and with the uniform horizontal line *BE* of [Figure 2](#) for different values of y_C . The example presented is for $y_C=0.2$ (curve 1), $y_C=0.5$ (curve 2), $y_C=0.8$ (curve 3) and $y_C=1.0$ (curve 4), while the main nozzle is taken for $y_C=1.0$. The intermediate contours of [Figure 6](#) will be used for determining the contours of the wall of the nozzle 3D after selecting the exit section. The example taken is for $M_E=300$ and $T_0=2000$ K.

[Figures 7](#), [8](#), [9](#), [10](#) and [11](#) represent respectively the variation of M , θ , T/T_0 , ρ/ρ_0 and P/P_0 on the intermediate contours of the nozzles walls of the [Figure 6](#). The axisymmetric nozzle has an inflection point considering the increase of θ from θ^* to θ_{Max} at the point of inflection and then decreases at $\theta=0.0$ at the exit of the nozzle.

[Figure 12](#) shows the shapes of the nozzle 3D for 16 selected sections. The shape 1 has a circular cross section. The shape 2 quadrilateral, the shape 3 square, the shape 4 is a half ellipse. The other 12 sections are arbitrarily chosen

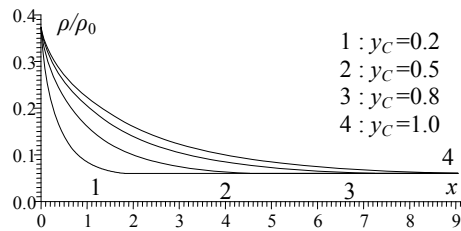


Fig. 10. Variation of ρ/ρ_0 on the contours of Figure 6.

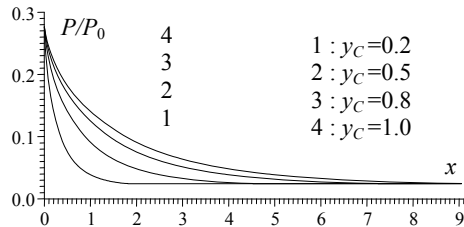


Fig. 11. Variation of P/P_0 on the contours of Figure 6.

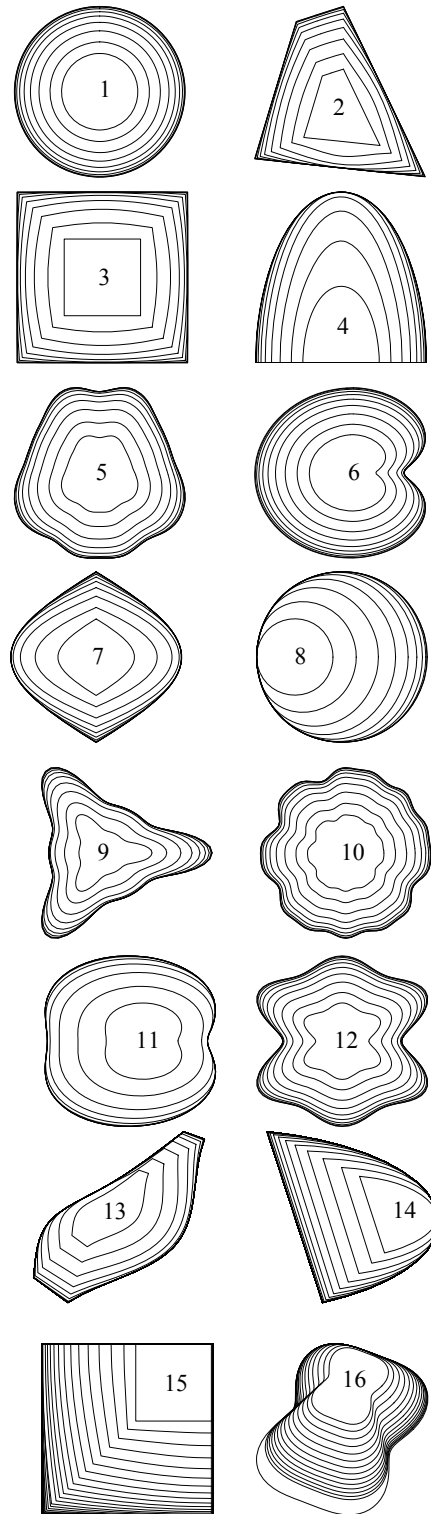


Fig. 12. Transverse contours of nozzle 3D for 16 exit section shapes when $M_E=3.00$ for $T_0=2000$ K.

to showcase the power of our numerical program. It should be noted that the shape of the throat section is identical to that chosen in the exit section. While the intermediate transverse shapes are totally different from that of the exit section except for the circular shape 1. This difference can be clearly seen on the 11 presented shapes. Our program can do any complex unsymmetrical section. It should be informed that the sections numbers 2, 4, 8, 13, 14, 15 and 16 do not contain the axis of symmetry of the parent nozzle.

The nozzles shown in Figure 12 all have the same longitudinal length $L=9.0504$ and the same ratio of the cross-sections $A_E/A^*=4.9958$ as those of the axisymmetric nozzle, considering that they are designed for $M_E=3.00$ and $T_0=2000$ K.

To make the comparison between the nozzle 3D and the axisymmetric from the point of view of the mass obtained and the thrust force delivered by the wall, it is necessary to design 3D shapes having the same critical cross-sectional area as the axisymmetric nozzle in addition to the preservation of M_E .

Figures 13, 14, 15, 16 and 17 show respectively the variation of M , θ , T/T_0 , ρ/ρ_0 and P/P_0 as a function of the transverse angle β of the nozzle 11 of Figure 12. The nozzle 11 has been taken as an example of an illustration to show that the physical parameters in the nozzle of arbitrary 3D section depend essentially on x , y and z if the Cartesian coordinates are used or on x , r and β if the cylindrical coordinate is used, with the exception of the circular section that the physical parameters do not depend on β . For this reason the axisymmetric nozzle is named by the quasi-three-dimensional nozzle, while the other sections are cases of three-dimensional flows. It is noted that in the throat section and the exit section of the nozzle 3D, the flow parameters do not depend on the angle β since the flow in both sections is unidirectional and even parallel. In the throat section $M=M^*$ which represents the value of M just after the expansion. Then through the nozzle there is an

abrupt increase in M between $M=1.0$ and $M=M^*$ at the same point A of Figure 1, then a progressive increase along the wall of the nozzle between $M=M^*$ at point A , to $M=M_E$ at the exit section (point E in Fig. 1).

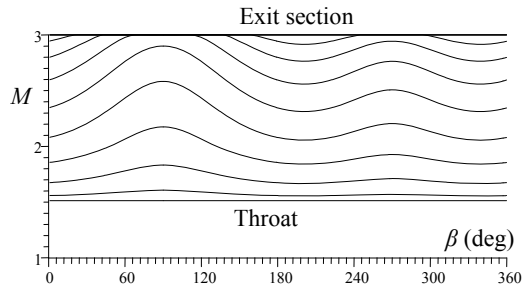


Fig. 13. Variation of M on the wall of the transverse sections of the nozzle 11 of Figure 12.

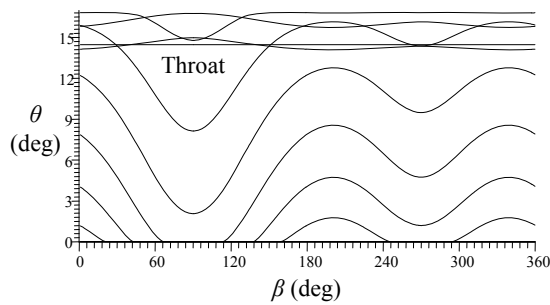


Fig. 14. Variation of θ on the wall of the transverse sections of the nozzle 11 of Figure 12.

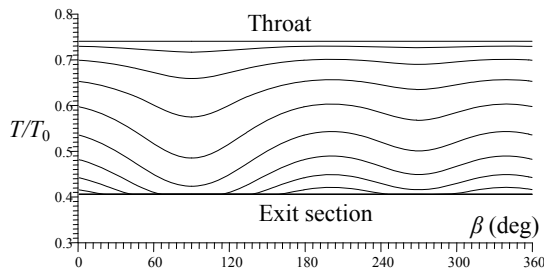


Fig. 15. Variation of T/T_0 on the wall of the transverse sections of the nozzle 11 of Figure 12.

Figure 14 shows that there is an increase in θ of $\theta = \theta^*$ at the point A to $\theta = \theta_{Max} > \theta^*$ at the inflection point which is in the vicinity of the throat, then decrease of θ from $\theta = \theta_{Max}$ of inflection point to $\theta = 0.0$ at the exit of the nozzle. On an intermediate section, the angle θ depends on the angle β of Figure 3. Only the circular section, the radius r of the cross-sections does not depend on the angle β . For the other non-circular sections, the radius r depends essentially on the angle β .

The results shown in Figures 13, 14, 15, 16 and 17 are valid for any shape of the exit section except for the circular section originally placed at a coordinate reference (nozzle 1 of Fig. 12).

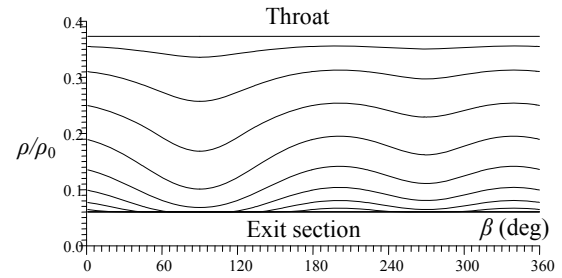


Fig. 16. Variation of ρ/ρ_0 on the wall of the transverse sections of the nozzle 11 of Figure 12.

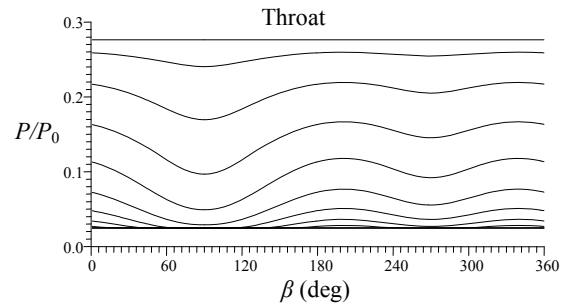


Fig. 17. Variation of P/P_0 on the wall of the transverse sections of the nozzle 11 of Figure 12.

7 Conclusion

A novel numerical effective and rapid method for the design of nozzle 3D of arbitrary exit section nozzles has been presented with respect to the axisymmetric nozzle applied to any type of nozzle in the case of HT or PG models. The following conclusions can be drawn from this work:

The design of the axisymmetric nozzle in the non-dimensional shape is necessary.

The choice of unsymmetrical exit section of the nozzle 3D is arbitrary.

The developed numerical program can process any natural gas. It suffices to add the function $C_p(T)$ and R of gas in question and to calculate the enthalpy $H(T)$. The application is made only for air.

An infinity of shape of the nozzles 3D according to the choice of the parameters M_E , T_0 , the shape of the exit section of the nozzle 3D and the gas can be obtained.

The most important of this method is that it is independent of the mesh results of the axisymmetric nozzle, which is not the case for the other methods used until now. This advantage has enabled us to considerably reduce the calculation time compared to the other methods currently used.

The nozzle 3D design by this method depends solely on the position and properties of the boundary points of the axisymmetric nozzle.

The calculation accuracy of this method depends on the design accuracy of the axisymmetric nozzle.

The execution time of this new method is decreased in a very considerable and important way and becomes negligible compared to the calculation time of the old design methods currently used.

The calculation of the current function is not necessary for this method for the determination of the coordinates of the points of the nozzle 3D, which is not the case for the other methods.

New contours of the nozzle 3D are found by this new method.

The design of the nozzle 3D can be done by this method using a simple non-powerful calculator with a very high discretization without interrupting the execution of numerical program.

The shape of the throat section of the nozzle 3D is identical with that of the exit section. These two sections verified the ratio of the critical sections of the axisymmetric nozzle.

Between the throat and the exit section, the nozzle 3D takes an arbitrary shape which is not identical to the throat and exit.

The axisymmetric form is used to compare the results found by our method.

As a very advanced technique to further improve the performances of the nozzle 3D, the nozzle 3D can be truncated in a section close to the exit section. In this case a gain of a portion of the mass is observed and in parallel a slight loss of thrust is noticed.

The flow in the axisymmetric nozzle depends on two variables (x and y) and does not depend on the angle β , while the flow in the nozzle 3D depends on three variables (x , r and β).

As a perspective, and in order to see the improvement of the performances of the nozzle 3D relative to the axisymmetric nozzle, the same cross-section of the throat between the two nozzles is maintained, in addition to fixing another parameter among the exit Mach number, the mass of the nozzle or the thrust coefficient. The two remaining parameters determine the performance of the new obtained shape.

A second problem in perspective consists in removing the uniform flow portion of the wall from the nozzle found for each contour in order to decrease in addition the mass of the nozzle 3D with the same Mach number performances and the delivered thrust force.

A third problem of improving the performances of the nozzle 3D is to place the exit section of the nozzle 3D in the uniform zone of the axisymmetric nozzle near the end of the Kernel region in order to decrease the maximum possible the undesirable uniform area.

Nomenclature

M	Mach number
x	Abscissa of a section of the nozzle
y, z	Position of a point in the section
r	Radius of a point in a section
θ	Flow angle deviation
ψ	Stream line function

β	Angle formed by the vector radius between two points of a section
$\Delta\theta$	Flow deviation step chosen in the expansion center A of Figure 1
Δ	Step progress of the characteristics on the uniform BE line in the transition zone
P	Pressure
T	Temperature
R	Thermodynamic constant of gas
H	Enthalpy
C_P	Specific heat at constant pressure
L	Length of the nozzle
γ	Specific heat ratio
ρ	Density
ε	Error
N_L	Number of longitudinal sections of the three dimensional nozzle
N_T	Number of points in a cross-section of the three dimensional nozzle
N_F	Number of points on the boundary of the axisymmetric nozzle
N_S	Number of mesh segments of axisymmetric nozzle
N_P	Number of mesh points of axisymmetric nozzle
HT	High Temperature
PG	Perfect Gas
MLN	Minimum Length Nozzle
MOC	Method Of Characteristics

Indices

0	Stagnation condition (combustion chamber)
*	Critical condition
E	Exit section
R	Right point of a segment
L	Left point of a segment

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